Milestone Review Flysheet

 Institution
 University of Alabama In Huntsville
 Milestone
 Preliminary
 Design Review

| First Stage (Both Stages Together or Single Stage) | | | | Second Stage (If Applicable) | | | | | |
|--|--|-------------------------------|---------------------------|------------------------------|--|------------------|-----------------|-----------|-----------|
| Vehicle Properties | | | | | Vehicle Properties | | | | |
| Total Length (in) | | | 122 in | | Total Length (in) | | | | |
| Diameter (in) | | | 4.7 | 7 in | | Diameter (in) | | | |
| Gross | Lift Off Weight | t (lb) | 29 | lb | (| Gross Weight (Ik | o) | | |
| Ai | irframe Materia | ıl | Carbon Fiber | | Airframe Material | | | | |
| | Fin Material | | Carbo | n Fiber | | Fin Material | | | |
| | | Notor Propertie | | | | | Motor Propertie | es | |
| Mot | or Manufacture | | | TI | Mo | tor Manufacture | | | |
| | tor Designation | ` ' | 7312M4770-P VMAX | | Motor Designation(s) | | • • | | |
| | Average Thrust | | 1362/1073 | | Max/Average Thrust (I | | | | |
| | al Impulse (lbf-s | | | 45 | Total Impulse (lbf-sec) | | • ' | | |
| | <u> </u> | tability Analysi | | | Ignition Altitude (ft) | | | | |
| Center of Pressure (in from nose) | | | | 5 in | Ignition Timing (From 1st Stage Burnou | | • | | |
| | Center of Pressure (III from nose) | | | .5in | Igniter Location | | | | |
| | Static Stability Margin | | | .5 | Stability Analysi | | is . | | |
| | | | | 5.6 | Center of Pressure (in from nose) | | | | |
| 11110 | Thrust-to-Weight Ratio Rail Size (in) | | | unistrut | Center of Pressure (in from nose) | | | | |
| | Rail Length (in) | | | ch tower | Static Stability Margin | | | | |
| - | Exit Velocity (ft | ·/s) | | ft/s | | ust-to-Weight R | | | |
| IXali | | Ascent Analysis | • | 11/3 | 1111 | ust-to-weight N | Ascent Analysis | | |
| Maxi | mum Velocity (| • | 1 | O ft/s | Max | rimum Velocity (| | | |
| - | mum Mach Nur | | | .7 | | imum Mach Nui | | | |
| | | | | . <i>1</i> ft/s^2 | | num Acceleratio | | | |
| | m Acceleration (1st Stage if M | | | 00 ft | | arget Apogee (f | | | |
| raiget Apogee | | | | 00 It | | | | a aution | |
| | | ery System Pro | | | | | ery System Pro | | |
| G C . | | rogue Parachut | | | Cartin | | Progue Parachut | te | |
| Configu | | | d, Semi-Hemispl | | | uration | | | |
| Siz | _ | 12 inch diameter | | | ze | | | | |
| Deployment V | | < 20 ft/s | | Deployment | | | | | |
| Terminal Ve | | 100 ft/s | | Terminal V | | | | | |
| Fabric | | Ripstop Nylon | | | Туре | | | | |
| Shroud Line | | Nylon Paratrooper Chord 500lb | | Shroud Line Material | | | | | |
| Shroud Line | | 36 in | | Shroud Line Length (in) | | | | | |
| Thread | | Not Determined | | Thread Type | | | | | |
| Seam ⁻ | | Not Determined | | Seam Type | | | | | |
| Recovery Harness Type | | Not Determined | | • | arness Type | | | | |
| Recovery Harne | ess Length (ft) | Not Determined | | Recovery Harness Length (ft) | | | | | |
| Harness/Airframe Interface | | Not Determined | | Harness/Airframe Interface | | | | | |
| | | Main Parachute | | | | | Main Parachute | | |
| | Configuration | | Round, Semi-Hemispherical | | | Configuration | | | |
| Siz | _ | 220 inch diameter | | | Size | | | | |
| Deployment Velocity (ft/s) | | 100 ft/s | | | Deployment Velocity (ft/s) | | | | |
| Terminal Velocity (ft/s) | | 7 ft/s | | | Terminal Velocity (ft/s) | | | | |
| Fabric Type | | Ripstop Nylon | | | Fabric Type | | | | |
| Shroud Line Material | | Nylon Paratrooper Chord 500lb | | | Shroud Line Material | | | | |
| Shroud Line Length (in) | | 144 in | | | Shroud Line Length (in) | | | | |
| Thread Type | | Not Determined | | | Thread Type | | | | |
| Seam Type | | Not Determined | | | Seam Type | | | | |
| Recovery Harness Type | | Not Determined | | Recovery H | arness Type | | | | |
| Recovery Harness Length (ft) | | Not Determined | | Recovery Harness Length (ft) | | | | | |
| Harness/Airfra | me Interface | | Not Determined | l | Harness/Airfr | ame Interface | | | |
| Vinatia France | Section 1 | Section 2 | Section 3 | Section 4 | Kingtin Free | Section 1 | Section 2 | Section 3 | Section 4 |
| Kinetic Energy | Booster/ | Nest: C | | | Kinetic Energy of Each | F: - C- | A.da.da.a | Ness C | |
| Of Each | | Nose Cone | Ī | | OI EGCII | Fin Can | Avionics Bay | Nose Cone | |
| of Each Section (ft-lbs) | Body Tube | | | | Section (ft-lbs) | | | | |

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| First Sta | ge (or Single Stage) | Second Stage (If Applicable) | | | |
|------------------------------|---------------------------------------|------------------------------|--|--|--|
| Recove | ry System Properties | Recovery System Properties | | | |
| | PerfectFlite SL-100 | | | | |
| Altimeter(s)/Timer(s) | PerfectFlight miniTimer4 | Altimeter(s)/Timer(s) | | | |
| (Make/Model) | | Make/Model | | | |
| | | | | | |
| | GPS Antenova M10382-AIUB (Locator) | | | | |
| Locators/Frequencies (Model- | Xbee PRO XSC-S3B 900MHz (Transmitter) | Locators/Frequencies (Model- | | | |
| Frequency-Power) | | Frequency-Power) | | | |
| | | | | | |
| Black Powder Charge Size | Not Determined | Black Powder Charge Size | | | |
| Drogue Parachute (grams) | Not Determined | Drogue Parachute (grams) | | | |
| Black Powder Charge Size | Not Determined | Black Powder Charge Size | | | |
| Main Parachute (grams) | Not Determined | Main Parachute (grams) | | | |

| | Payloads |
|-----------|---|
| | Overview |
| Mandatory | Landing Hazard Detection System - A Video system with object detection algorithms to identify possible hazards in the landing area. |
| Payload | |
| 3.1 | |
| | Overview |
| Optional | Dielectrophoresis in Micro Gravity - Study of using Electric Fields to manipulate liquid fuels in micro gravity. |
| Payload 1 | |
| | |
| 3.3.1.1 | |
| | Overview |
| Optional | Supersonic Effects on Vehicle Coatings - Apply different surface products to the vehicle airframe and to observe the effects of supersonic flight |
| Payload 2 | in a post flight analysis. |
| | |
| 3.3.2.1 | |

| Test Plans, Status, and Results | | |
|---------------------------------|----------------|--|
| Ejection Charge Tests | Not Determined | |
| Sub-scale Test Flights | Not Determined | |
| Full-scale Test Flights | Not Determined | |

Additional Comments

Prometheus will feature a 4th payload in support of the NanoLaunch Project which includes a variety of gyroscopes, accelerometers, and pressure sensors to provide meaningful data in an attempt to characterize vehicle aerodynamic coefficients during transonic flight. The vehicle will have an induced pitched to determine pitching moment coefficient.